

THE VEHICLE

THE SATELLITE



www.ilslaunch.com

PROTON HISTORY

- Lead designer was Vladimir Chelomei, who designed it with the intention of creating a powerful rocket for both military payloads and as a high-performance ICBM. The program was changed, and the rocket was developed exclusively for launching spacecraft.
- First named UR-500, but adopted the name "Proton," which also was the name of the first three payloads launched.
- Proton launched Russian interplanetary missions to the Moon, Venus, Mars, and Halley's Comet.
- Proton launched the Salyut space stations, the Mir core segment and both the Zarya (Dawn) and Zvezda (Star) modules for today's International Space Station.
- First commercial Proton launch — 9 April 1996, ASTRA 1F.

PROTON DESCRIPTION

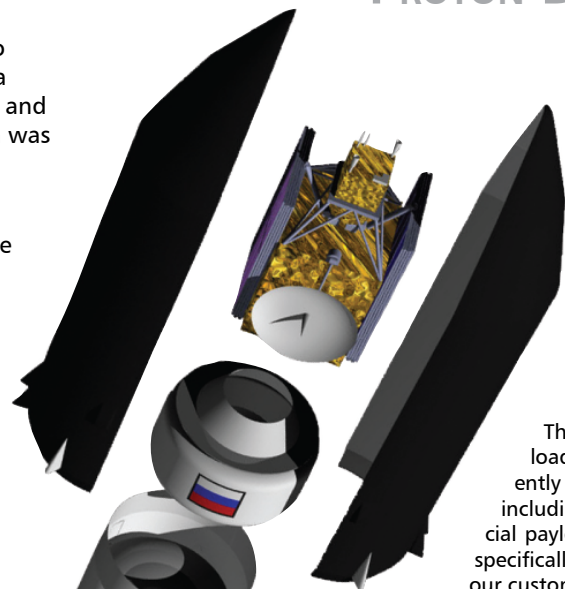
TOTAL HEIGHT
58.2 m (191 ft)

GROSS LIFTOFF WEIGHT
702,000 kg
(1,547,000 lb)

PROPELLANT
UDMH and N_2O_4

INITIAL LAUNCH
16 July 1965
Proton-1 Spacecraft

PAYLOAD FAIRINGS
There are multiple payload fairing designs presently qualified for flight, including standard commercial payload fairings developed specifically to meet the needs of our customers.



BREEZE M UPPER STAGE

The Breeze M is powered by one pumped gimbaled main engine that develops thrust of 19.6 kN (4,400 lbf). The Breeze M is composed of a central core and a jettisonable additional propellant tank. Inert mass of the stage at liftoff is approximately 2,665 kg (5,875 lb). The quantity of propellant carried is dependent on specific mission requirements and is varied to maximize mission performance. The Breeze M is controlled by a closed loop, triple-redundant guidance system.

PROTON BOOSTER

The Proton booster is 4.1 m (13.5 ft) in diameter along its second and third stages, with a first stage diameter of 7.4 m (24.3 ft). Overall height of the three stages of the Proton booster is 42.3 m (138.8 ft).

THIRD STAGE

Powered by one RD-0213 engine, this stage develops thrust of 582 kN (131,000 lbf), and a four-nozzle vernier engine that produces thrust of 31 kN (7,000 lbf).

SECOND STAGE

Of conventional cylindrical design, this stage is powered by three RD-0210 engines plus one RD-0211 engine and develops a vacuum thrust of 2.5 MN (570,000 lbf).

FIRST STAGE

The first stage consists of a central tank containing the oxidizer surrounded by six outboard fuel tanks. Each fuel tank also carries one of the six RD-276 engines that provide first stage power. Total first stage sea-level thrust is approximately 10.0 MN (2,250,000 lbf) with a vacuum-rated level thrust of 11.0 MN (2,500,000 lbf).

The Proton and the Breeze M are built by Khronichev State Research and Production Space Center.



SATELLITE OPERATOR

Ciel Satellite Group
www.cielsatellite.ca

SATELLITE MANUFACTURER

Thales Alenia Space
www.thalesaleniaspace.com

PLATFORM

Spacebus 4000 C4

SEPARATED MASS

5,561 kg

SATELLITE DESIGN LIFE

16 Years

MISSION

Ciel II is the largest and most technologically advanced Spacebus 4000 satellite built to date by Thales Alenia Space. The Ciel II satellite introduces a new generation of Canadian telecommunications satellites offering the highest quality digital television services to consumers in the Canadian and North American markets. Ciel II will be located at 129° West Longitude.

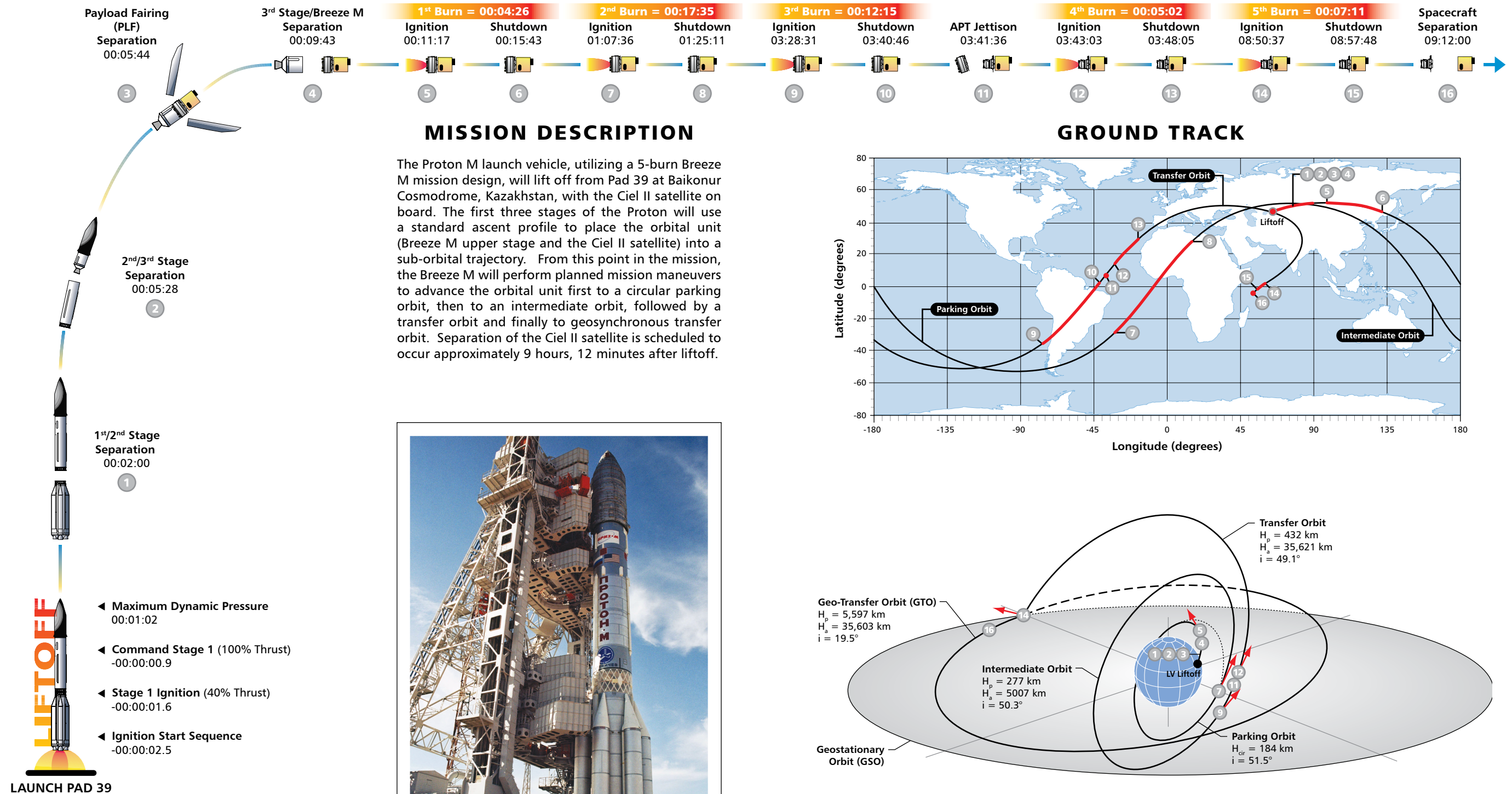


Ciel II

MISSION OVERVIEW

- 49th Proton launch for ILS
- 3rd Launch of a Spacebus 4000 on a Proton
- 1st dedicated satellite for Ciel Satellite Group

THE MISSION

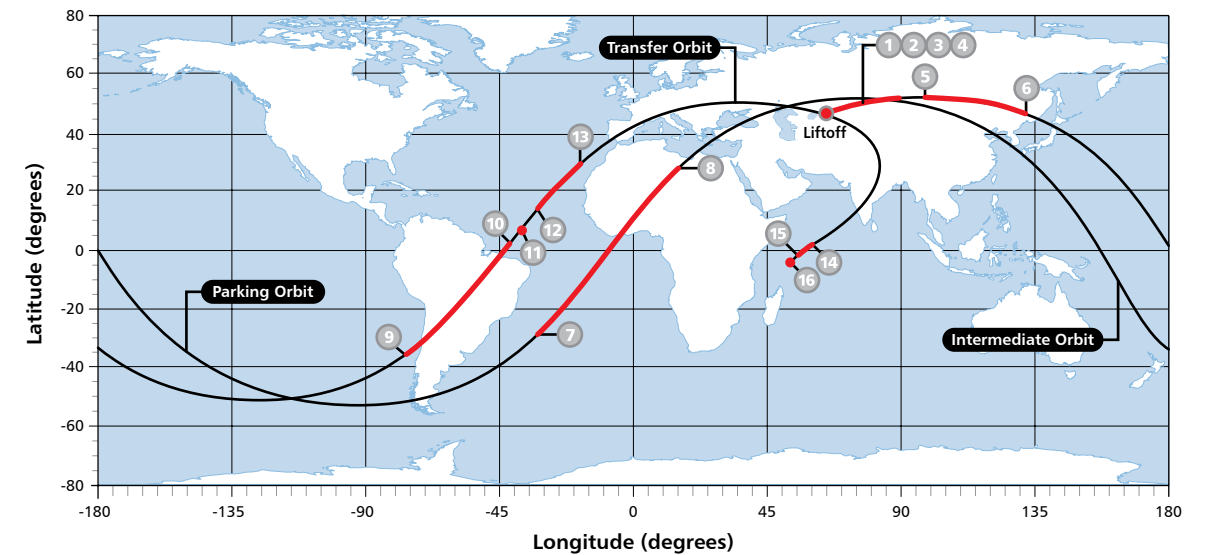


MISSION DESCRIPTION

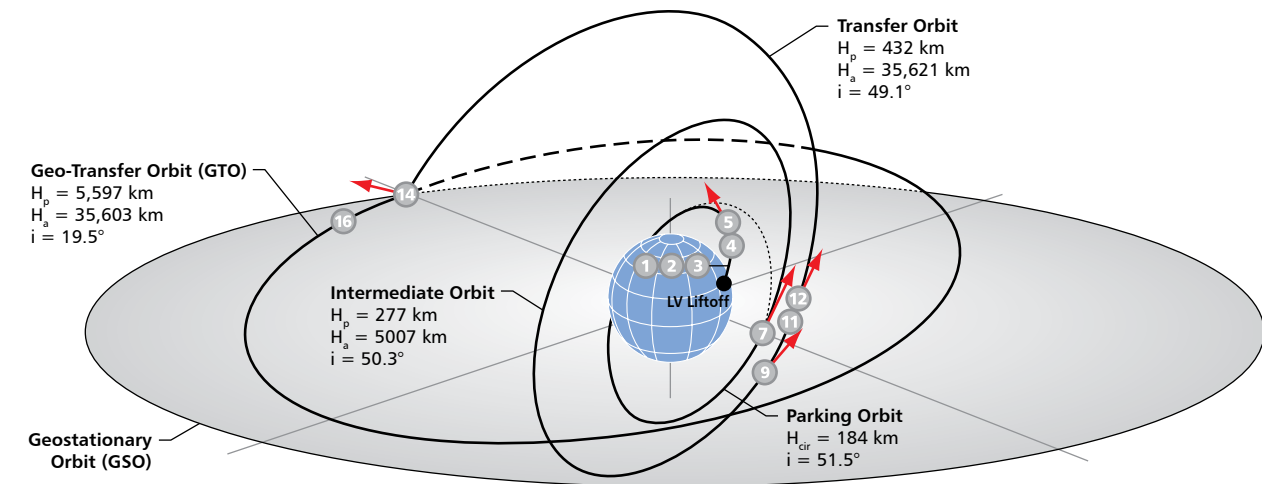
The Proton M launch vehicle, utilizing a 5-burn Breeze M mission design, will lift off from Pad 39 at Baikonur Cosmodrome, Kazakhstan, with the Ciel II satellite on board. The first three stages of the Proton will use a standard ascent profile to place the orbital unit (Breeze M upper stage and the Ciel II satellite) into a sub-orbital trajectory. From this point in the mission, the Breeze M will perform planned mission maneuvers to advance the orbital unit first to a circular parking orbit, then to an intermediate orbit, followed by a transfer orbit and finally to geosynchronous transfer orbit. Separation of the Ciel II satellite is scheduled to occur approximately 9 hours, 12 minutes after liftoff.



GROUND TRACK



ORBIT INSERTION



ASCENT PROFILE

PROTON M ON PAD 39